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STATEMENT REGARDING GOVERNMENT RIGHTS

The government has certain rights to this invention under

Department of Defense Contract No. N00140-91-C-2793.--

IN THE CLAIMS:

Cancel claims 8, 12, 15, 18 and 24-26, without prejudice to submission of the subject matter thereof in a continuation application.

Rewrite reissue claims 4, 9-11, 16, 19-23, 27, 42, 43, 30 and 32, as follows:

4. A turbomachinery blade for a gas turbine engine fan comprising a plurality of blades mounted for rotation about a fan axis with neighboring blades forming passages for a working medium gas, wherein:

the blade has a configuration enabling the fan to rotate at speeds providing supersonic flow velocities over the blade in at least a portion of each passage causing the formation of a shock in the gas adjacent an inner wall of a case forming an outer boundary for the working medium gas flowing through the passages;

the blade has a leading edge with an inner region ending at an inward boundary of an intermediate region and a tip region